

**SHUTTLE CRITICAL ITEMS LIST - ORBITER**

SUBSYSTEM : ATMOSPHERIC REVIT.      PMEA NO 06-LA -1601 -1      REV: 07/08/88

ASSEMBLY : AIRLOCK					CRIT. FUNC: 1R
P/N RI : MC250-0004-0012					CRIT. HDW: 2
P/N VENDOR: 2763-0001-9		VEHICLE	102	103	104
QUANTITY : 2		EFFECTIVITY:	X	X	X
: TWO ON INNER HATCH		PHASE(S):	PL	LO	CO X DO LS

		REDUNDANCY SCREEN:	A-PASS	B-N/A	C-PASS
PREPARED BY:		APPROVED BY:	APPROVED BY (NASA)		
DES S. CASTILLO		DES <i>[Signature]</i>	SSM	<i>[Signature]</i>	
REL D. RISING		REL <i>[Signature]</i>	REL	<i>[Signature]</i>	
QE W. SMITH		QE <i>[Signature]</i>	QE	<i>[Signature]</i>	

ITEM:  
EQUALIZATION VALVE, CABIN/AIRLOCK HATCH

FUNCTION:  
PROVIDES FOR EQUALIZING PRESSURE ACROSS THE INNER HATCH. EACH VALVE OPERATES INDEPENDENTLY WITH POSITIVE DETENTS AT TWO FLOW POSITIONS. VALVE CAN BE ACTUATED FROM EITHER SIDE OF HATCH.

FAILURE MODE:  
INABILITY TO OPEN, RESTRICTED FLOW

CAUSE(S):  
VIBRATION, MECHANICAL SHOCK, CORROSION, CONTAMINATION, PHYSICAL BINDING/JAMMING

EFFECT(S) ON:  
(A) SUBSYSTEM (B) INTERFACES (C) MISSION (D) CREW/VEHICLE  
(A, B) LOSS OF ONE REDUNDANT VALVE TO REPRESSURIZE AIRLOCK.  
(C, D) NO EFFECT.  
(E) FUNCTIONAL CRITICALITY EFFECT - SECOND ASSOCIATED FAILURE (LOSS OF OTHER EQUALIZATION VALVE) COULD RESULT IN LOSS OF EVA CREWMEN. SCREEN 'B' IS N/A BECAUSE REDUNDANT VALVE IS STANDBY.

DISPOSITION & RATIONALE:  
(A) DESIGN (B) TEST (C) INSPECTION (D) FAILURE HISTORY (E) OPERATIONAL USE

(A) DESIGN  
UNIT IS FLANGE MOUNTED WITH A SINGLE SILASTIC-675 SILICONE RUBBER O-RING WHICH COMPENSATES FOR ROUGHNESS OF FLANGE PREVENTING EXTERNAL LEAKAGE. HOUSING IS FABRICATED OF A356.0-T61 ALUMINUM ALLOY.

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(B) TEST

QUALIFICATION TESTS FOR 100 MISSION LIFE: ACCELERATION OF 5 G FOR FIVE MINUTES PER AXIS. SINUSOIDAL VIBRATION - 5 TO 15 HZ AT +/- 0.25 G PER AXIS. RANDOM VIBRATION - 0.09 G<sup>2</sup>/HZ FOR 48 MINUTES PER AXIS. DESIGN SHOCK - 20 G PER AXIS. THERMAL VACUUM/THERMAL CYCLE - WITH VALVE CLOSED AND CAP ON, UNIT EXPOSED TO 120 TO 130 F AND VACUUM OF 1 X 10<sup>-7</sup> TORR FOR 24 HOURS. LOW/HIGH TEMPERATURE CYCLE - HELD AT -40 TO -50 F FOR 3 HOURS AND AT +120 TO 130 F FOR 3 HOURS. OPERATING LIFE - OPERATION OFF/NORMAL/EMERGENCY POSITIONS WITH 15 PSIG APPLIED FOR 800 CYCLES. LEAKAGE MONITORED DURING OR AFTER THESE TESTS LIMITED TO 5 SCCM MAX.

ACCEPTANCE TEST - PROOF PRESSURE 25 PSIG GN<sub>2</sub>, WITH VALVE OPEN AND CLOSED. LEAK CHECK AT 15 PSIG, 5 SCCM MAX - VALVE OPEN AND CLOSED AND REVERSE LEAKAGE.

OMRAD - AIRLOCK VALVES OPENING AND CLOSING TORQUES VERIFIED EVERY TWO FLIGHTS. CONTINGENCY TEST AT 15 PSIG AFTER LRU REPLACEMENT AND CONTINGENCY TEST AT 3.2 PSID IN VEHICLE. GROSS LEAKAGE TEST AT 2 PSIG BEFORE EACH FLIGHT.

(C) INSPECTION

RECEIVING INSPECTION

MATERIALS VERIFIED AT RECEIVING INSPECTION. ALUMINUM HOUSING CASTINGS ARE HYDROSTATIC PROOF PRESSURE TESTED AT 32 PSID.

CONTAMINATION CONTROL

CORROSION PROTECTION PROVISIONS AND CONTAMINATION CONTROL PLAN VERIFIED BY INSPECTION. CLEANLINESS LEVELS OF 200A AND 100 ML RINSE TEST VERIFIED.

ASSEMBLY/INSTALLATION

MANUFACTURING PROCESSES, INSTALLATION AND ASSEMBLY VERIFIED BY INSPECTION. TORQUES VERIFIED BY INSPECTION. DIMENSIONAL CHECKS PERFORMED BY INSPECTION. INSPECTION PERFORMS MIPs FOR CONCENTRICITY AND PERPENDICULARITY.

NONDESTRUCTIVE EVALUATION

ALUMINUM HOUSING CASTINGS ARE X-RAYED AND DYE PENETRANT INSPECTED TO DETECT CRACKS, VERIFIED BY INSPECTION.

CRITICAL PROCESSES

PASSIVATED PARTS AND HEAT TREATMENT VERIFIED BY INSPECTION. MECHANICAL SOLDERING OF DEBRIS SCREEN VERIFIED BY INSPECTION. ANODIZATION OF ALUMINUM PARTS VERIFIED BY INSPECTION.

TESTING

ATP VERIFIED BY INSPECTION.

HANDLING/PACKAGING

PARTS PROTECTION VERIFIED BY INSPECTION.

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(D) FAILURE HISTORY  
NO FAILURE HISTORY.

(E) OPERATIONAL USE  
CREW WILL USE THE REDUNDANT VALVE FOR REPRESSURIZATION OF AIRLOCK.